










Method of producing an element of composite material

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Inventor: PODESTA EMANUELE (IT)
Applicant: ELASIS SIST RICERCA FIAT NEL M (IT)
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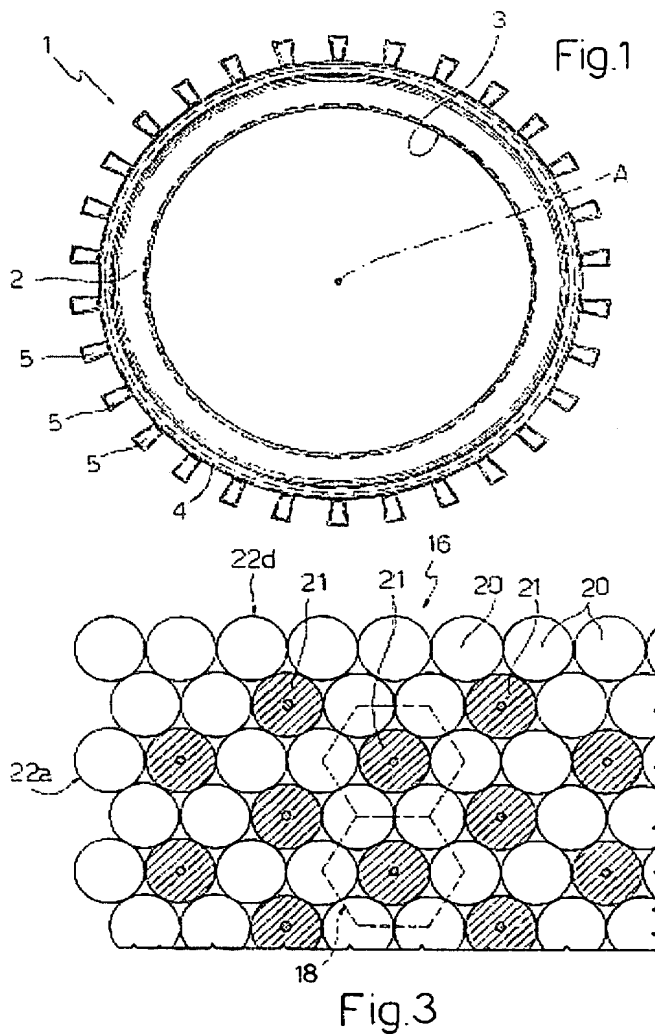
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 GB1155708
 EP0667207
 US5460774
 EP0846550
 EP0490629
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Abstract of EP1099774

A method of producing an element of composite material (1), including the steps of forming a first distribution of first elements (20) defining the matrix of the element of composite material (1); forming a second distribution of second elements (21) defining the reinforcing structure of the element of composite material (1); and compacting the first and second elements (20, 21) to obtain a distribution of the reinforcing structure inside the matrix; the first elements being metal wires (20); and the step of forming the first distribution including the step of assigning each second element (21) an orderly distribution of metal wires (20).



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(72) Inventor: **Podesta, Emanuele**

72100 Brindisi (IT)

(74) Representative: **Franzolin, Luigi et al**

STUDIO TORTA S.r.l.,

Via Viotti, 9

10121 Torino (IT)

(71) Applicant: **ELASIS SISTEMA RICERCA FIAT NEL
MEZZOGIORNO Società Consortile per Azioni
80038 Pomigliano d'Arco, Napoli (IT)**

(54) Method of producing an element of composite material

(57) A method of producing an element of composite material (1), including the steps of forming a first distribution of first elements (20) defining the matrix of the element of composite material (1); forming a second distribution of second elements (21) defining the reinforcing structure of the element of composite material (1);

and compacting the first and second elements (20, 21) to obtain a distribution of the reinforcing structure inside the matrix; the first elements being metal wires (20); and the step of forming the first distribution including the step of assigning each second element (21) an orderly distribution of metal wires (20).

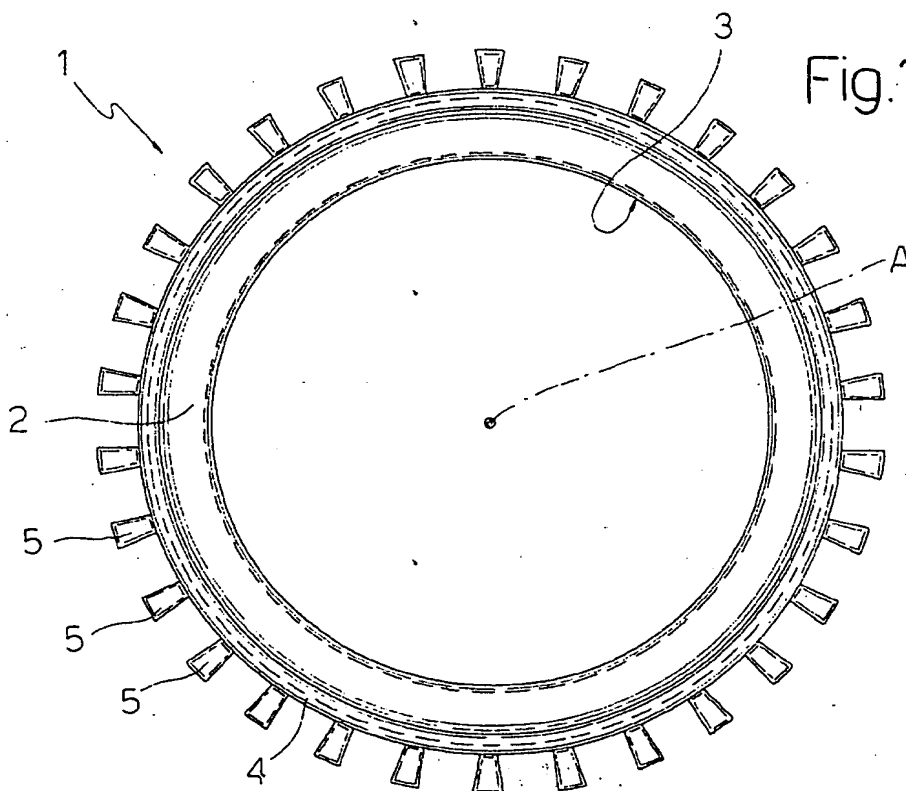


Fig.1

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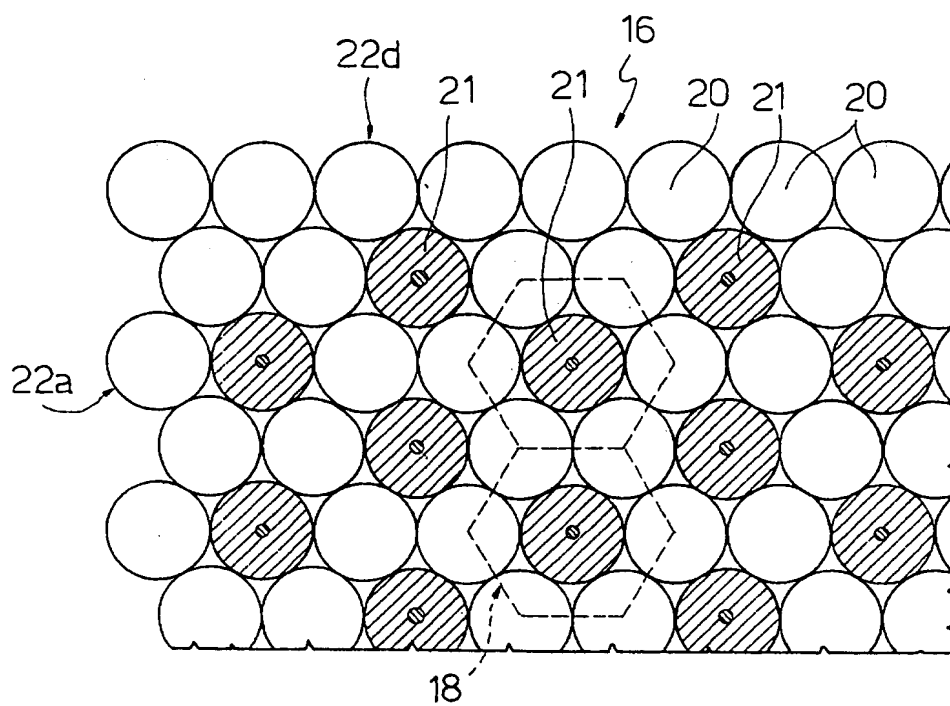


Fig.3

Description

[0001] The present invention relates to a method of producing elements of composite material, in particular, circular-geometry elements such as countershafts, turbine and compressor disks for turbomachines, etc.

[0002] As is known from Italian Patent Application n. T096A000979 filed on 3 December 1996 by FIATAVIO S.p.A, composite-material elements of the above type are produced by forming a number of disks, each formed by winding a continuous reinforcing fiber about an axis to form a flat spiral; stacking the disks with the interposition of respective spacer sheets of metal material; and axially compacting the stack to form a metal matrix in which the various spirals of reinforcing fibers are embedded.

[0003] The physical characteristics of such composite-material elements depend mainly on the distribution of the reinforcing fibers inside the metal matrix; and the extent to which the fibers are distributed evenly depends on the extent to which the turns in each disk are equally spaced a predetermined distance apart, and the extent to which the freedom of movement of the various turns is restricted, especially at the compacting stage.

[0004] For which reason, the turns of reinforcing fiber are locked in place with respect to one another by fastening wires wound about each turn and extending spokefashion with respect to the axis of the spiral.

[0005] More specifically, the turns are equally spaced a given distance apart by forming, alongside formation of the spiral, a further two flat spirals of spacer wire, which are removed from the spiral of reinforcing fiber once the fastening wires are wound about the turns.

[0006] The method described briefly above involves several drawbacks.

[0007] In particular, producing composite-material elements using disks of reinforcing material and metal spacer sheets of given thicknesses means it is impossible to obtain any given desired distribution of the reinforcing fibers inside the metal matrix.

[0008] Moreover, the above method comprises various fairly complex, and therefore fairly high-cost, operations (weaving the spirals of reinforcing wire separately and fastening the relative turns; stacking the disks of ceramic material and spacer sheets; and placing the stacks inside a final container to form the composite-material elements).

[0009] In the case of a titanium metal matrix, the spacer sheets are not easy to procure in the form required by the methods described, i.e. of constant 0.1 mm thickness, and call for various dedicated machining operations (cutting, grinding, welding, etc.) which further increase the already high cost involved.

[0010] Finally, the fastening wires must be made of inert material, with respect to both the metal matrix and the reinforcing fibers.

[0011] It is an object of the present invention to provide a method of producing an element of composite

material, designed to eliminate in a straightforward, low-cost manner the aforementioned drawbacks typically associated with known methods.

[0012] According to the present invention, there is provided a method of producing an element of composite material comprising a metal matrix and a reinforcing structure, said method comprising the steps of:

- forming a first distribution of first elements defining said matrix;
- forming a second distribution of second elements defining said reinforcing structure; and
- compacting said first and second elements to obtain a distribution of said reinforcing structure inside said matrix;

characterized in that said first elements are metal wires; and in that said step of forming said first distribution comprises the step of assigning each said second element an orderly distribution of said metal wires.

[0013] A preferred, non-limiting embodiment of the present invention will be described by way of example with reference to the accompanying drawings, in which:

[0014] Figure 1 shows a front view of an element of composite material formed in accordance with the present invention;

[0015] Figure 2 shows an axial section of a supporting body with a ring of composite material, from which the Figure 1 element is formed using the method according to the present invention;

[0016] Figure 3 shows a larger-scale view of a detail of the Figure 2 ring;

[0017] Figures 4 to 9 show partial axial sections of successive operating steps in the formation of the Figure 1 element according to the method of the present invention;

[0018] Figure 10 shows the Figure 3 detail following application of the method according to the present invention.

[0019] Number 1 in Figure 1 indicates as a whole an element of composite material formed using the method according to the present invention - in the example shown, a rotary member, such as a compressor disk for turbomachines, to which the following description refers purely by way of example.

[0020] Element 1 is of circular annular shape with an axis of symmetry A, and comprises a central portion 2 in the form of a flat disk and defining a through hole 3 of axis A, and a substantially cylindrical peripheral portion 4 projecting axially in both directions with respect to central portion 2 and supporting externally a number of projecting radial blades 5.

[0021] More specifically, central portion 2 is made of a composite material defined by a matrix of metal material - in the example shown, titanium alloy - and by a reinforcing structure of ceramic material - in the example shown, silicon carbide - and is coated externally with a

thin layer of metal or so-called "skin", preferably of titanium alloy.

[0022] Peripheral portion 4, on the other hand, is made entirely of metal material, advantageously the same material as the matrix of central portion 2.

[0023] Element 1 is formed by preparing and then compacting a toroidal base structure 6 (Figure 6) of axis A.

[0024] Structure 6 is formed from a substantially annular main body 7 (Figures 2, 4-9) comprising a through hole 8 of axis A defining hole 3 of element 1, and a disk-shaped portion 9, from a flat end surface 10, perpendicular to axis A, of which projects axially a cylindrical-tubular portion 11 having an outside diameter smaller than the outside diameter of disk-shaped portion 9.

[0025] Hole 8 is defined at portions 9 and 11 by respective cylindrical surfaces 12, 13 having different diameters and connected to each other by a flat intermediate surface 14 perpendicular to axis A and extending along an extension of end surface 10. More specifically, cylindrical surface 12 is larger in diameter than cylindrical surface 13.

[0026] Main body 7 also comprises an annular projection 15, of axis A, projecting inside hole 8 from intermediate surface 14 and having a right-triangular section with the hypotenuse facing cylindrical surface 13.

[0027] Base structure 6 is formed as follows.

[0028] First of all, a first distribution of metal wires 20 defining the metal matrix of element 1, and a second distribution of fibers 21 of ceramic material defining the reinforcing structure of element 1 are positioned coaxially on main body 7.

[0029] An important characteristic of the present invention is that the first distribution is formed by assigning each fiber 21 an orderly distribution of metal wires 20. Wires 20 and fibers 21 together define a composite-material ring 16 (Figure 2) woven on a known winding machine not shown. In the example shown, wires 20 and fibers 21 are annular with a circular section (Figure 3) and are made respectively of titanium alloy and silicon carbide.

[0030] More specifically, ring 16 is positioned coaxially about tubular portion 11 of main body 7, and rests on end surface 10 of disk-shaped portion 9.

[0031] Wires 20 and fibers 21 are advantageously combined in a weave pattern (Figure 3) in which two wires 20 are interposed between each pair of fibers 21. More specifically, in the weave pattern, each fiber 21 is surrounded by six wires 20 forming the vertices of a hexagon, and occupies the barycenter of the hexagon.

[0032] Ring 16 is defined externally by a radially outer and radially inner cylindrical lateral surface 22a, 22b, and by two opposite flat annular end surfaces 22c, 22d; which surfaces 22a, 22b, 22c, 22d are made exclusively of metal wires 20 for ensuring, after the compacting step, the structural continuity of ring 16, main body 7 and the other metal parts of structure 6 described in detail later on.

[0033] Wires 20 and fibers 21 have the same diameter and together define a number of hexagonal base cells 18 (shown by the dash lines in Figure 3); and each base cell 18 is defined by a central fiber 21 and by respective 120° angular portions of the six wires 20 surrounding central fiber 21, so that the volume of the reinforcing structure is 33% that of the matrix.

[0034] Structure 6 is completed by fitting main body 7 coaxially with two annular closing elements 23, 24 (Figures 4 and 5) and a cover 25 (Figure 6), which, together with main body 7, define a closed seat for ring 16.

[0035] With particular reference to Figures 4-9, closing element 23 is the same axial height as tubular portion 11 of main body 7, while the axial height of closing element (or piston ring) 24 equals the difference between the axial heights of tubular portion 11 and ring 16.

[0036] Closing element 23 is fitted onto the radially outer surface 22a of ring 16 so as to rest on end surface 10 of disk-shaped portion 9 of main body 7; and, similarly, closing element 24 is inserted between tubular portion 11 of main body 7 and closing element 23 so as to rest on end surface 22d of ring 16, on the opposite side to disk-shaped portion 9.

[0037] Cover 25 comprises a circular, annular, disk-shaped wall 28, from the radially inner and outer peripheral edges of which project respective concentric inner and outer cylindrical walls 29, 30.

[0038] Cover 25 is assembled by positioning disk-shaped wall 28 facing respective free axial ends of closing elements 23, 24 and tubular portion 11 of main body 7, and by inserting cylindrical wall 29 inside hole 8 so that the end rests on projection 15, and by fitting cylindrical wall 30 on the outside of closing element 23 so that the end rests on a peripheral annular shoulder 31 of disk-shaped portion 9 of main body 7 (Figure 6).

[0039] Cover 25 is then fixed to main body 7 by spot welding the portions contacting projection 15 and shoulder 31.

[0040] At this point, the air inside structure 6 is extracted using a known molecular pump (not shown) and a known muffle furnace (not shown) for heating structure 6 to a temperature of about 600°C.

[0041] The resulting structure 6 is compacted in a conventional autoclave (not shown) for HIPping (Hot Isostatic Pressing) processing with automatic temperature and pressure control.

[0042] At the first stage, lasting about two hours, the temperature of the autoclave, initially at ambient conditions, is increased to the superplasticity temperature of the titanium alloy - in the example described, about 900°C.

[0043] The temperature in the autoclave is then maintained constant long enough to enable the entire mass defining structure 6 to reach a uniform temperature. This period of time - two hours on average - is calculated bearing in mind that heat transmission at this stage is slowed down by the absence of air inside structure 6, and by the fact that the contact area between wires 20

of surfaces 22a, 22b, 22c, 22d of ring 16 and main body 7 is extremely small and therefore permits very little heating by conduction of wires 20. At the same time, the pressure inside the environment housing structure 6 and defined by the autoclave is increased to such a threshold value - in the example described, 900 Kg/cm² - as to permanently deform disk-shaped wall 28 of cover 25 in a direction parallel to axis A (Figure 7). More specifically, disk-shaped wall 28 of cover 25 flexes so as to come to rest on closing element 24, which in turn presses against composite-material ring 16 to act as a pressure equalizer and transmitter. Once disk-shaped wall 28 of cover 25 is so deformed as to enable closing element 24 to axially stress composite-material ring 16, metal wires 20 are deformed so as to fill the gaps formerly present between wires 20 and fibers 21. At this stage, composite-material ring 16 contracts along axis A, while the position of fibers 21 with respect to axis A remains constant to ensure uniform distribution of the reinforcing structure inside the metal matrix.

[0044] At this point, the pressure inside the autoclave is increased further to such a threshold value - in the example shown, about 1300 Kg/cm² - as to collapse the whole of structure 6, which is also compacted crosswise to axis A (Figure 9). More specifically, cylindrical walls 29, 30 of cover 25 adhere respectively to a radially outer surface of closing element 24 and to surface 13 defining hole 8, while composite-material ring 16 adheres along metal peripheral surfaces 22a, 22b, 22c, 22d to disk-shaped and tubular portions 9, 11 of main body 7 and to closing elements 23 and 24.

[0045] The compacted structure 6 is then cooled by so reducing the temperature and pressure as to minimize the residual stress produced in the portion derived from composite-material ring 16 by the different coefficients of thermal expansion of the metal matrix and reinforcing fibers 21.

[0046] The portion of element 1 derived from ring 16 assumes the Figure 10 configuration, in which fibers 21 are evenly distributed inside the metal matrix, are equally spaced in a direction perpendicular to axis A, and are separated by varying distances in a direction parallel to axis A.

[0047] Finally, the compacted structure 6 may be subjected to mechanical machining or similar to obtain the finished contour of element 1. In particular, blades 5 are formed from the part of compacted structure 6 derived from disk-shaped portion 9 of main body 7.

[0048] Using metal wires 20 to form the matrix of composite-material element 1 therefore provides, by appropriately selecting the diameter of wires 20 and fibers 21, for obtaining any desired distribution of the reinforcing structure inside the metal matrix.

[0049] In particular, by appropriately selecting the type of distribution of metal wires 20 relative to each reinforcing fiber 21, e.g. by adopting the hexagonal distribution described previously, the freedom of movement of fibers 21 can be limited during compaction to maintain

the positions of fibers 21 with respect to axis A.

[0050] Moreover, unlike known methods, the method described provides for forming composite-material element 1 by weaving wires 20 and fibers 21 directly onto parts (main body 7) eventually forming part of the metal matrix of element 1, thus eliminating the need for producing separate disks of reinforcing wire, fastening the turns of each disk, the long, complicated process of stacking the disks with respective metal spacer sheets in between, and placing the stacks inside containers for producing elements 1.

[0051] The spacer sheets, which are particularly expensive when titanium-based, and the work involved in preparing the sheets may therefore be eliminated with considerable saving.

[0052] Finally, contraction of structure 6 at the compacting stage is less than that of stacks of ceramic disks and metal spacer sheets using the known methods described previously.

[0053] Clearly, changes may be made to the method described and illustrated herein without, however, departing from the scope of the accompanying Claims.

[0054] In particular, reinforcing fibers 21 may be made of different materials, including metal.

[0055] Main body 7, closing elements 23, 24 and cover 25 may be made of different metal materials from each other and from the material of wires 20.

[0056] Finally, once formed, composite-material ring 16 may even be extracted from structure 6 and used to form different composite-material elements.

Claims

1. A method of producing an element of composite material (1) comprising a metal matrix and a reinforcing structure, said method comprising the steps of:

- forming a first distribution of first elements (20) defining said matrix;
- forming a second distribution of second elements (21) defining said reinforcing structure; and
- compacting said first and second elements (20, 21) to obtain a distribution of said reinforcing structure inside said matrix;

characterized in that said first elements are metal wires (20); and in that said step of forming said first distribution comprises the step of assigning each said second element (21) an orderly distribution of said metal wires (20).

2. A method as claimed in Claim 1, characterized in that said second elements are reinforcing fibers.

3. A method as claimed in Claim 2, characterized in

- that said assigning step comprises the step of preparing a woven element (16) by placing at least one said metal wire (20) alongside each said reinforcing fiber (21).
4. A method as claimed in Claim 3, characterized in that said step of preparing said woven element (16) comprises the step of interposing at least two said metal wires (20) between each pair of adjacent said reinforcing fibers (21).
 5. A method as claimed in Claim 3 or 4, characterized in that said step of preparing said woven element (16) comprises the step of surrounding each said reinforcing fiber (21) with six said metal wires (20) forming the vertices of a hexagon.
 6. A method as claimed in Claim 5, characterized in that said step of preparing said woven element (16) comprises the step of positioning each said reinforcing fiber (21) at the barycenter of the hexagon defined by said metal wires (20) about the reinforcing fiber (21).
 7. A method as claimed in any one of Claims 3 to 6, characterized in that said step of preparing said woven element (16) comprises the step of forming respective boundary surfaces (22a, 22b, 22c, 22d) of the woven element (16) using exclusively said metal wires (20).
 8. A method as claimed in any one of Claims 3 to 7, characterized in that said metal wires (20) and said reinforcing fibers (21) are annular; and in that said step of preparing said woven element (16) is performed by placing said metal wires (20) and said reinforcing fibers (21) about a toroidal main body (7) made of metal material.
 9. A method as claimed in Claim 8, characterized by comprising the step of forming a base structure (6) by fitting covering means (23, 24, 25) of metal material onto said main body (7) to close said woven element (16) between said main body (7) and the covering means (23, 24, 25).
 10. A method as claimed in Claim 9, characterized in that said main body (7) and said covering means (23, 24, 25) define, at the end of said compacting step, respective peripheral portions of said element of composite material (1); and in that said woven element (16) defines, at the end of said compacting step, a core of said element of composite material (1).
 11. A method as claimed in Claim 9 or 10, characterized in that said compacting step comprises the steps of:
 - placing said base structure (6) in an environment of controllable temperature and pressure conditions;
 - varying the temperature of said environment so as to bring said metal wires (20), said main body (7) and said covering means (23, 24, 25) uniformly to a superplasticity temperature; and
 - varying the pressure of said environment to deform said metal wires (20) and fill any gaps between the metal wires (20) and said reinforcing fibers (21) of said woven element (16), and to cause said main body (7) and said covering means (23, 24, 25) to adhere to one another and to said metal wires (20) of said boundary surfaces (22a, 22b, 22c, 22d) of said woven element (16).
 12. A method as claimed in any one of the foregoing Claims, characterized in that said metal wires are made of a titanium-alloy-based material.
 13. A method as claimed in any one of Claims 2 to 12, characterized in that said reinforcing fibers are made of ceramic material.
 14. A method as claimed in Claim 13, characterized in that said reinforcing fibers are made of a silicon-carbide-based material.
 15. A rotary member (1) made of composite material and comprising a structure of metal material (4) and a reinforcing element (2, 16) of composite material; characterized in that said reinforcing element (2, 16) is obtained from an orderly distribution of metal wires (20) and reinforcing fibers (21), and has respective boundary surfaces (22a, 22b, 22c, 22d) made exclusively from said metal wires (20) and connected integrally by compaction to said structure of metal material (4).

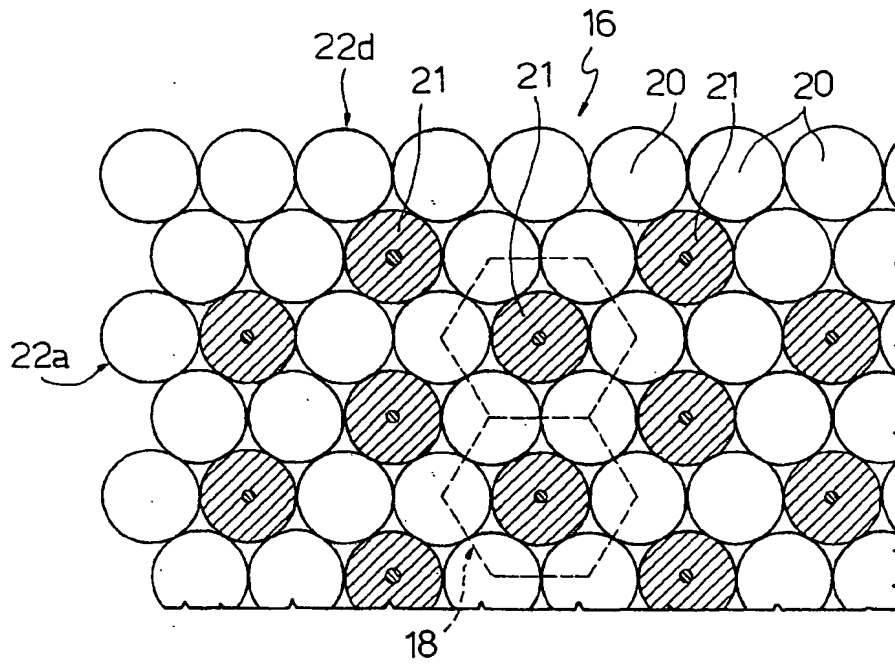


Fig.3

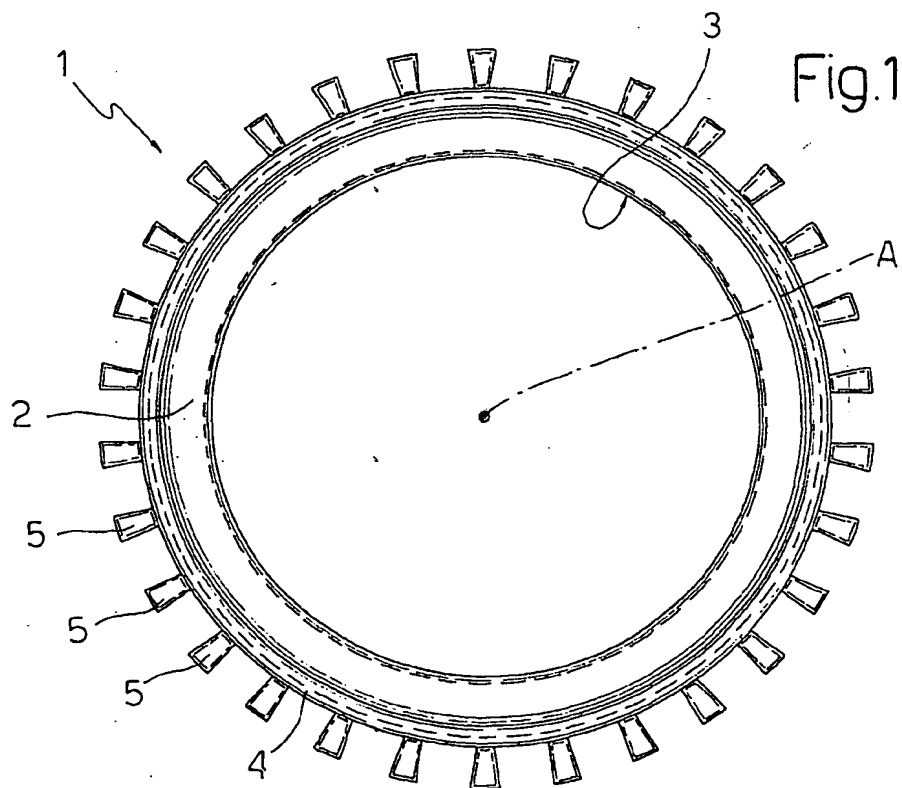


Fig.1

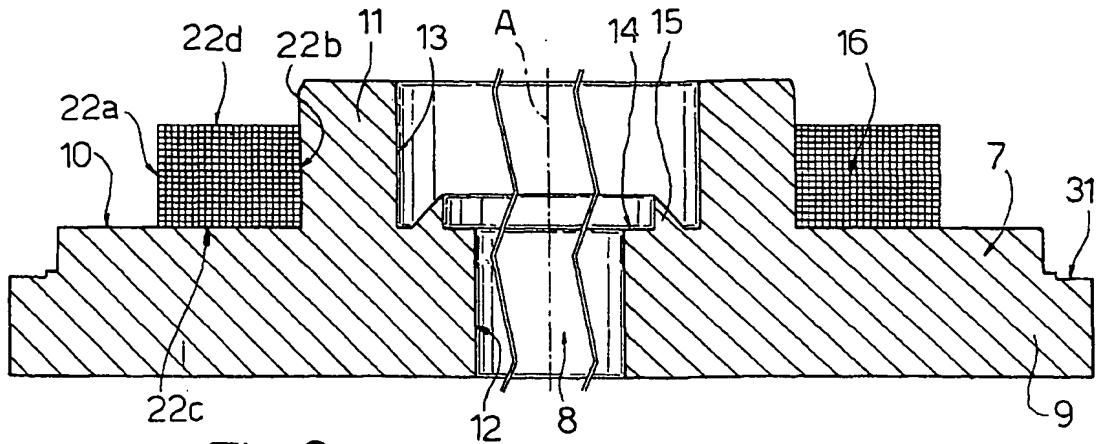


Fig. 2

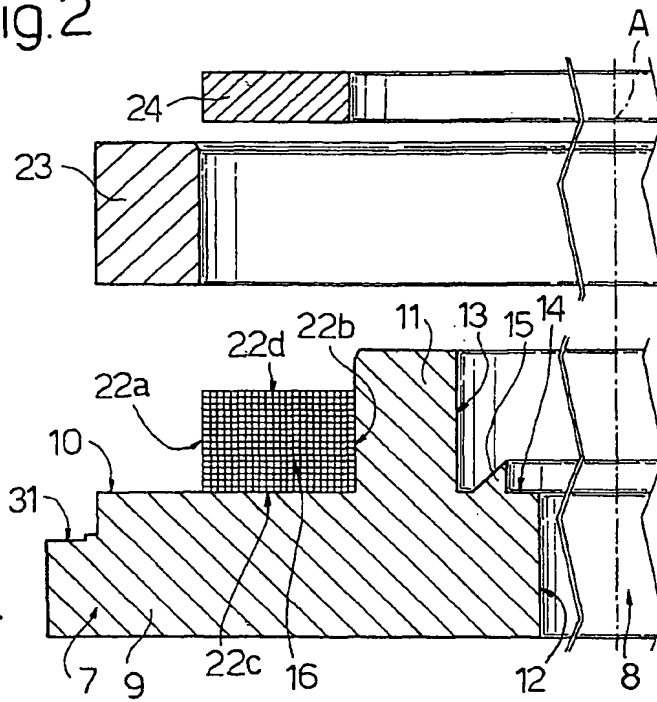


Fig. 4

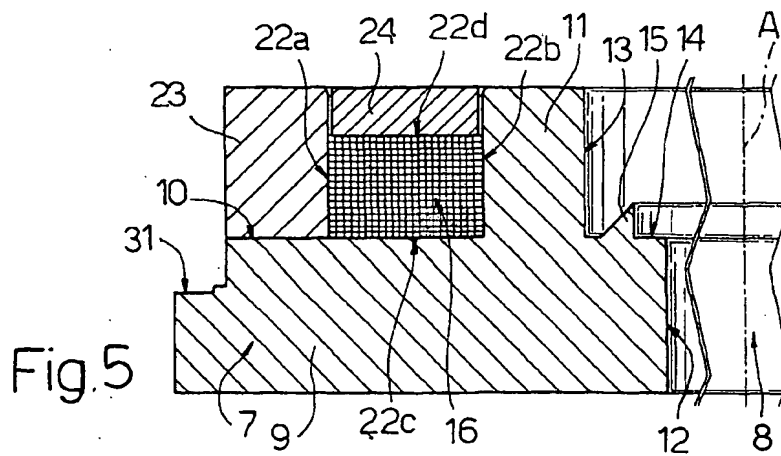
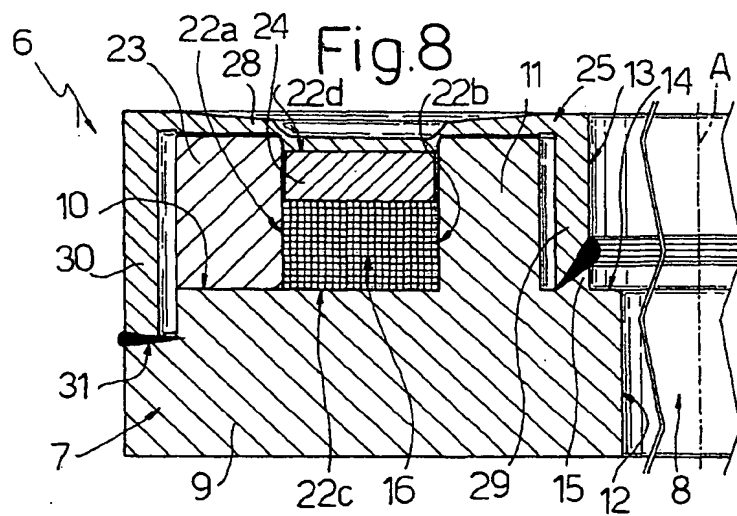
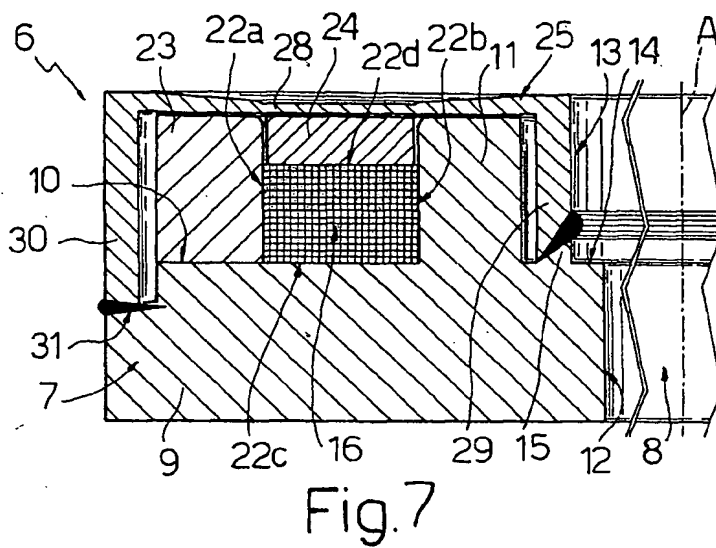
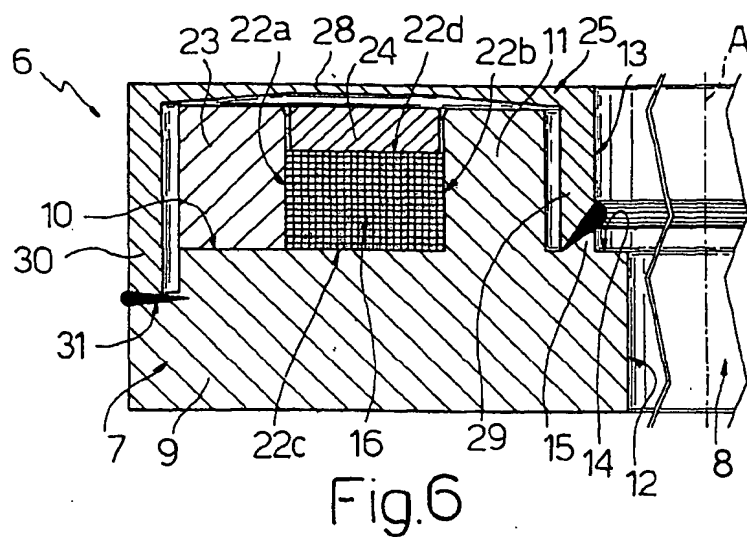


Fig. 5



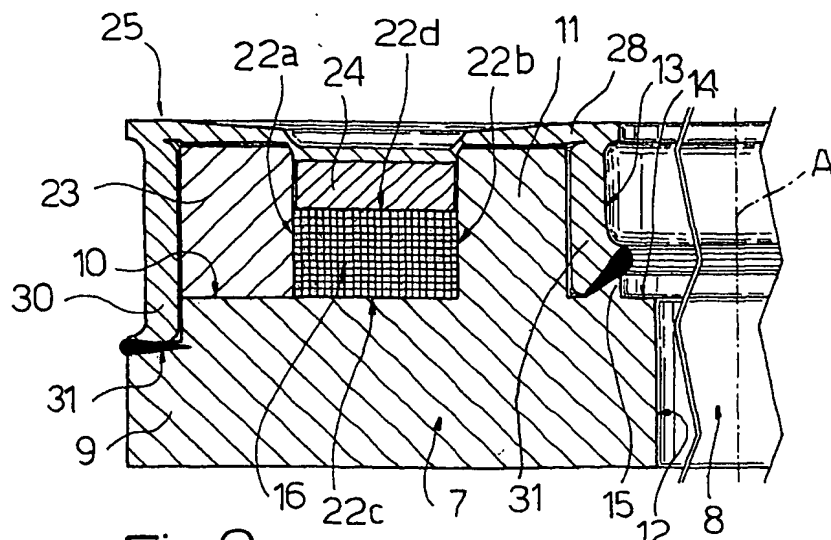


Fig. 9

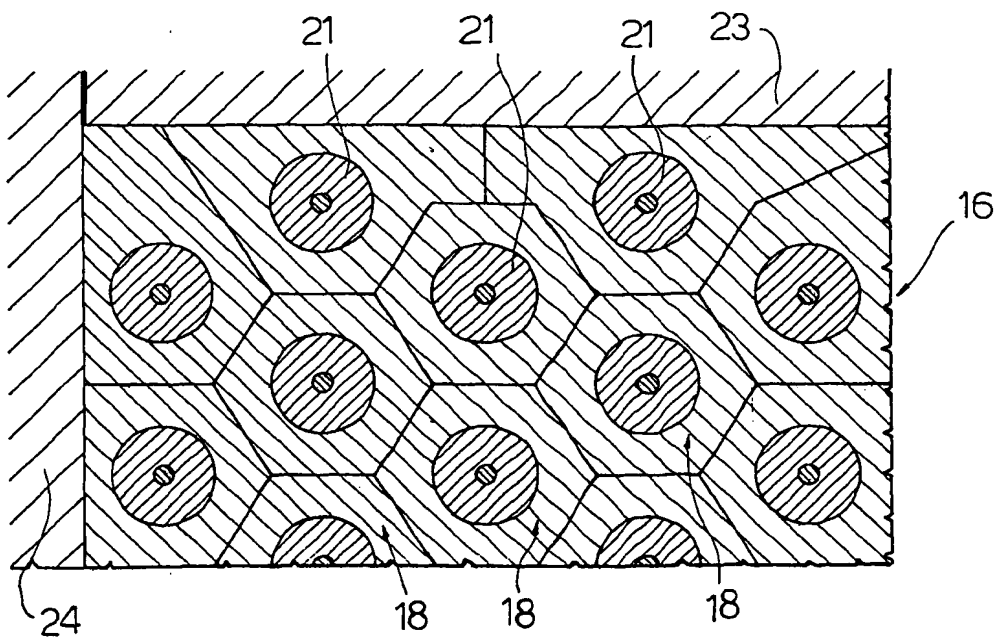


Fig. 10



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EUROPEAN SEARCH REPORT

Application Number
EP 99 83 0693

DOCUMENTS CONSIDERED TO BE RELEVANT				
Category	Citation of document with indication, where appropriate, of relevant passages	Relevant to claim	CLASSIFICATION OF THE APPLICATION (Int.Cl.7)	
X	GB 1 155 708 A (HANS HEUMANN) 18 June 1969 (1969-06-18)	1-10, 15	C22C1/09	
Y	* page 3, line 75 - line 93; claims 1-3, 10; figures 3, 11, 18, 23 *	11-14		
Y	EP 0 667 207 A (SNECMA) 16 August 1995 (1995-08-16) * column 9, line 23 - line 29; claim 1 *	11-14		
A	US 5 460 774 A (BACHELET ERIC) 24 October 1995 (1995-10-24)			
D, A	EP 0 846 550 A (FIATAVIO SPA) 10 June 1998 (1998-06-10)			
A	EP 0 490 629 A (AVCO CORP) 17 June 1992 (1992-06-17)			
A	PONZI C: "METAL MATRIX COMPOSITE FABRICATION PROCESSES FOR HIGH PERFORMANCE AEROSPACE STRUCTURES" COMPOSITES MANUFACTURING, GB, BUTTERWORTH SCIENTIFIC, GUILDFORD, SURREY, vol. 3, no. 1, 1 January 1992 (1992-01-01), pages 32-42, XP000287416 ISSN: 0956-7143			TECHNICAL FIELDS SEARCHED (Int.Cl.7)
A	US 3 864 807 A (SCHNEIDER FRIEDRICH ET AL) 11 February 1975 (1975-02-11)			C22C
A	WO 98 11265 A (MINNESOTA MINING & MFG) 19 March 1998 (1998-03-19)			
A	EP 0 831 154 A (ROLLS ROYCE PLC) 25 March 1998 (1998-03-25)			
The present search report has been drawn up for all claims				
Place of search THE HAGUE		Date of completion of the search 11 April 2000	Examiner Gregg, N	
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11-04-2000

Patent document cited in search report		Publication date	Patent family member(s)	Publication date
GB 1155708	A	18-06-1969	BE 682999 A	23-12-1966
			CH 464654 A	
			DE 1527553 A	11-12-1969
			NL 6608862 A	27-12-1966
EP 0667207	A	16-08-1995	FR 2715883 A	11-08-1995
			DE 69501836 D	30-04-1998
			DE 69501836 T	20-08-1998
			JP 2790241 B	27-08-1998
			JP 8066779 A	12-03-1996
			US 5544805 A	13-08-1996
US 5460774	A	24-10-1995	FR 2713663 A	16-06-1995
EP 0846550	A	10-06-1998	IT T0960979 A	03-06-1998
EP 0490629	A	17-06-1992	US 5431984 A	11-07-1995
			DE 69126285 D	03-07-1997
			DE 69126285 T	06-11-1997
			JP 4332639 A	19-11-1992
			US 5427304 A	27-06-1995
			US 5337940 A	16-08-1994
US 3864807	A	11-02-1975	DE 2059179 A	08-06-1972
			CH 538317 A	15-08-1973
			FR 2116467 B	02-04-1976
			GB 1369048 A	02-10-1974
WO 9811265	A	19-03-1998	AU 4343397 A	02-04-1998
			EP 0938592 A	01-09-1999
EP 0831154	A	25-03-1998	US 5946801 A	07-09-1999

EPO FORM P/0499

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